

IN THE CLAIMS

Please cancel claims 3 and 4 and amend claims 1, 6, 12, 13, 15 and 16 as follows:

1. (Currently Amended) A gas turbine combustor comprising a combustion chamber, diffusive combustion nozzles which inject fuel into air in said combustion chamber to form a diffusive combustion flame, an annular premixing flow passage formed by outer and inner walls in said combustion chamber and premixing nozzles disposed in said premixing flow passage for injecting fuel therein to mix with air to form a premixed gas, which flows out into said combustion chamber to form a premixing flame, characterized in that

a plurality of said premixing nozzles are mounted in spaced relationship in said premixing flow passage;

a plurality of spaced openings are formed in said outer wall through which air flows to mix with fuel from said premixing nozzles and form a swirling flow with respect to each of said premixing nozzles along an axial direction of said premixing flow passage; and

said openings are disposed in a circumferential direction whereby one opening is provided for each two adjacent nozzles at an intermediate position between the two adjacent nozzles in the circumferential direction.

2. (Previously Presented) A gas turbine combustor according to claim 1 wherein said swirling flows for said two adjacent nozzles rotate in opposite directions.

Claims 3-4. Cancelled

5. (Previously Presented) A gas turbine combustor according to claim 1, characterized in that each of said openings is configured in such a manner that a width of each opening in a circumferential direction varies along an axial direction thereof.

6. (Currently Amended) A gas turbine combustor ~~according to claim 5, characterized in that~~ comprising a combustion chamber, diffusive combustion nozzles which inject fuel into air in said combustion chamber to form a diffusive combustion flame, an annular premixing flow passage formed by outer and inner walls in said combustion chamber and premixing nozzles disposed in said premixing flow passage for injecting fuel therein to mix with air to form a premixed gas, which flows out into said combustion chamber to form a premixing flame, characterized in that

a plurality of said premixing nozzles are mounted in spaced relationship in said premixing flow passage;

a plurality of spaced openings are formed in said outer wall through which air flows to mix with fuel from said premixing nozzles and form a swirling flow with respect to each of said premixing nozzles;

said openings are disposed in a circumferential direction whereby one opening is provided for each two adjacent nozzles;

each of said openings is configured in such a manner that a width of each opening in a circumferential direction varies along an axial direction thereof; and

each of said openings is configured in ~~a planform trapezoid~~ a generally triangular shape in such a manner either that the opening broadens in a main air stream direction prior to air flowing into the premixing flow passage or that the opening decreases in the main air stream direction prior to air flowing into the premixing flow passage.

7-11. (Cancelled)

12. (Currently Amended) A gas turbine combustor using a premixing device comprising a plurality of premixing nozzles which are arranged in a circumferential direction and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into a combustion chamber, characterized in that one air flow inlet for every adjacent two premixing nozzles is provided at an intermediate position

between the two adjacent nozzles in the circumferential direction so that a swirling flow is formed with respect to each of said adjacent two premixing nozzles along and around each of said adjacent two premixing nozzles.

13. (Currently Amended) A gas turbine combustor using a premixing device comprising a plurality of premixing nozzles which are arranged in a circumferential direction and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into a combustion chamber, characterized in that one air flow inlet for every adjacent two premixing nozzles is provided at an intermediate position between the two adjacent nozzles in the circumferential direction so that a swirling flow is formed with respect to each of said two adjacent premixing nozzles along and around each of said adjacent two premixing nozzles and said swirling flows for said two adjacent nozzles rotate in opposite directions.

14. (Cancelled)

15. (Currently Amended) A premixing method for a gas turbine combustor comprising a plurality of premixing nozzles which are arranged in a circumferential direction and form a premixing combustion flame by injecting premixed gas formed by

premixing fuel and air into a combustion chamber, characterized in that air flows in from air flow inlets, with one of said air inlets being provided for every adjacent two premixing nozzles at an intermediate position between the two adjacent nozzles in the circumferential direction, and a swirling flow is formed along and around each of said adjacent two premixing nozzles.

16. (Currently Amended) A premixing method for a gas turbine combustor comprising a plurality of premixing nozzles which are arranged in circumferential direction and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into a combustion chamber, characterized in that air flows in from air flow inlets, with one of said inlets being provided for every adjacent two premixing nozzles at an intermediate position between the two adjacent nozzles in the circumferential direction, and a swirling is formed along and around said adjacent two premixing nozzles, and said swirling flows for said two adjacent nozzles rotate in opposite directions.

17. (Cancelled)